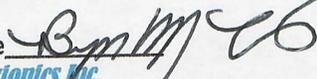


THE **adlog**<sup>TM</sup> AIRCRAFT  
MAINTENANCE  
RECORDKEEPING  
SYSTEM

1964 BEECH D95A N8831M  
TD-574  
AVIONICS

**AVIONICS  
MAINTENANCE  
RECORDS**



DATE	AIRFRAME TIME IN SERVICE	AVIONICS TIME IN SERVICE	DESCRIPTION OF WORK PERFORMED— SIGNATURE & CERTIFICATE NO. OF PERSON PERFORMING WORK
			<div data-bbox="344 389 1395 562" style="border: 1px solid black; padding: 5px;"> <p>N8831M <span style="float: right;">7/10/23</span></p> <div style="display: flex; justify-content: space-between;"> <div data-bbox="354 488 516 517">Hobbs: 4944.0</div> <div data-bbox="766 389 964 562" style="text-align: center;">  <p>SMART Avionics Inc. FAA CRS ZV8R725X 186 Airport Road Marietta, PA 17547</p> </div> <div data-bbox="1247 488 1386 548" style="text-align: right;"> <p>Work Order RS23-148</p> </div> </div> </div> <p>Removed System 60-1 AP Control, Roll Computer and Pitch Computer. Installed Genesys System 3100 Autopilot Computer Control P/N 01326-07-02-000 S/N 01326-2244-00002 in pilot instrument panel. Fabricated new wiring harness using mil spec wire M22579 wire and milspec M27500 shielded wire in accordance with Genesys installation instructions ST-964-II-0001 rev 3. System was interfaced to existing pitch servo, existing trim servo, existing roll servo and existing Yaw servo. System was interfaced to the Aspen EFD 1000 Max Instrument system for Attitude information, Airdata information and navigation information. System installation approved by Genesys STC SA09755DS.</p> <p>Removed Pitch servo, Roll servo, Yaw servo and Pitch Trim servo for overhaul. Installed Pitch Servo P/N 0108-P4 S/N 4016A. Installed Roll Servo P/N 0106-R9 S/N 4032A. Installed Yaw Servo P/N 0106-Y10 S/N 4038A. Installed Pitch Trim Servo P/N 0106-T2 S/N 15675. System was ground tested and performed as specified with no adverse effects to other aircraft systems.</p> <p>All workmanship was accomplished in accordance with the guidelines of AC 43.13-1B chapters 7 sections 2, 4 chapter 10 section 1, chapter 11 sections 1 thru 12 &amp; chapter 12 sections 1, 2 and AC43.13-2B chapters 1 section 100, chapter 2 sections 203, 207, 209 and chapter 3 sections 304, 306, 307, 308 and 310. Electrical load does not exceed 80% of aircraft electrical system capacity. The aircraft weight and balance and the equipment list were updated to reflect this change.</p> <div style="text-align: right; margin-top: 20px;">               Benjamin Travis              Repairman 2774970         </div>
			<div data-bbox="344 1377 1104 1769" style="border: 1px solid black; border-radius: 10px; padding: 10px; margin: 10px auto; width: 80%;"> <p>Registration: <u>N8831M</u>      Date <u>6-20-25</u></p> <p>I certify the Transponder test and Inspection required by FAR 91.413 IAW Part 43, App F and the Altimeter and Static System test required by FAR 91.411 IAW Part 43, App E have been performed. The Altimeter was tested to <u>20,000</u> Feet and the results are on file under Work Order <u>RS25-115</u></p> <p>Signature   <u>SMART Avionics Inc.</u>                      FAA Repair Station ZV8R725X      Hobbs: <u>5070.2</u></p> </div>